

AM Payload: NASA Sounding Rocket

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Experiment Overview

- Deliver AM experiments onto a NASA sounding rocket for experimental evaluation
- Payload limited in power, but recovered
- Opportunity to expose AM parts to combined mechanical environments
- Sounding rocket flew on March 22nd from NASA Wallops

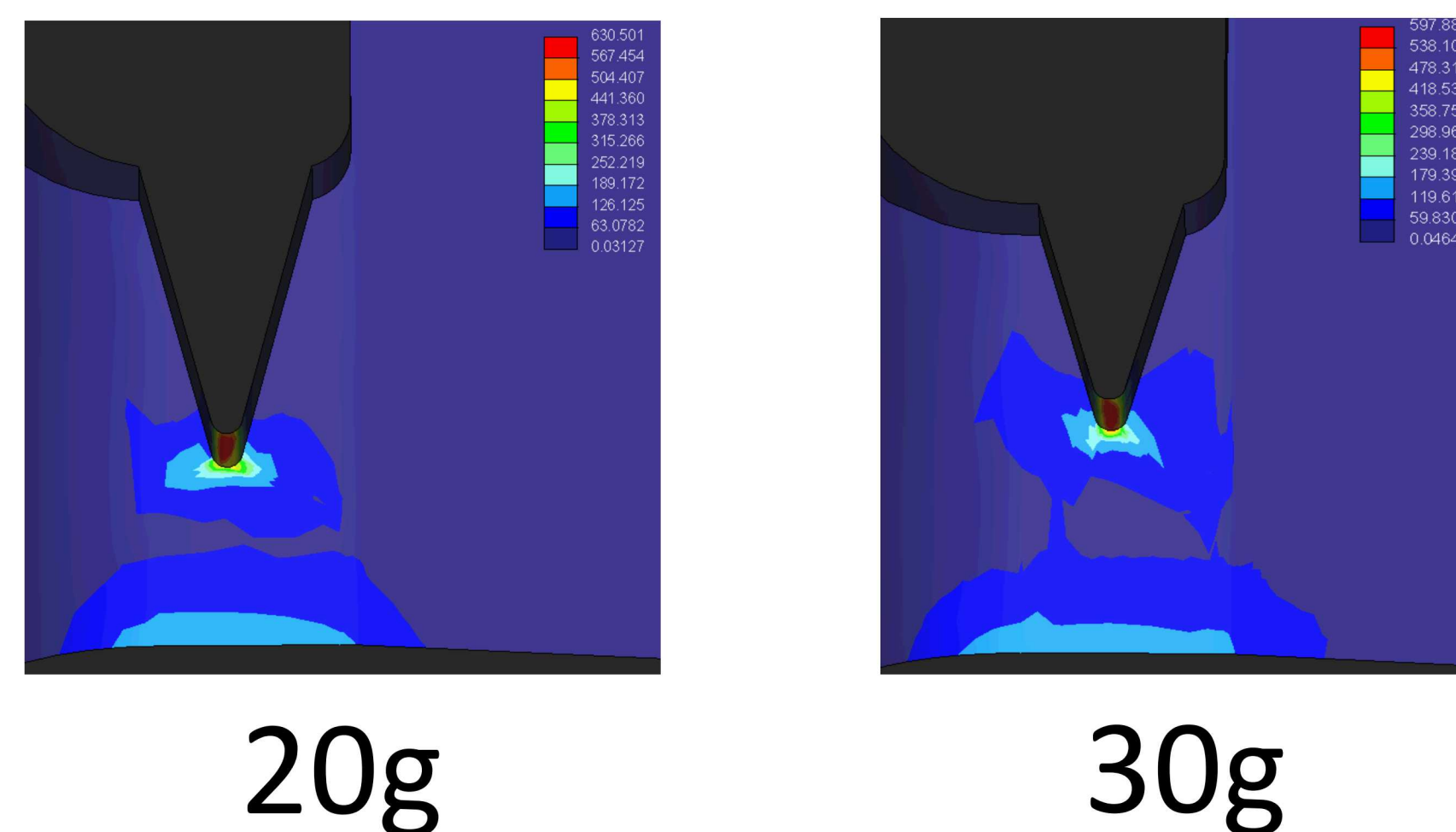
Experiment Details

- Two sets of four cantilever beams designed to fail at specific accelerations
- Cantilevers fabricated via LENS process with final machining to get tolerances in critical regions
- Beams CT scanned prior to integration to look at shape and pores in critical region



Modeling Efforts

- Notch geometry designed such that launch stresses are greater than ultimate tensile strength
- Hand calculations used to optimize notch geometry
- FEA conducted to validate hand calculations



Input Variables

$$\alpha = 30 \text{ deg} \quad t = 0.075 \text{ in} \quad h = 0.17 \text{ in} \quad D = 0.3 \text{ in} \quad r = 0.01 \text{ in} \quad \rho = 7700 \frac{\text{kg}}{\text{m}^3}$$

$$t_{\text{notch}} = 0.25 \text{ in} \quad \theta = 7.5 \text{ deg} \text{ in}$$

$$\frac{h}{r} = 17 \quad A = t_{\text{notch}} \cdot D = (4.839 \cdot 10^{-6}) \text{ m}^2 \quad d = D - h = 0.13 \text{ in}$$

Calculate Stress Factors

$$C_1 = 2.966 + 0.502 \cdot \left(\frac{h}{r}\right) - 0.009 \cdot \left(\frac{h}{r}\right)^2 \quad C_2 = 8.023 + 1.253 \cdot \left(\frac{h}{r}\right) - 0.020 \cdot \left(\frac{h}{r}\right)^2$$

$$C_3 = -6.475 - 1.126 \cdot \left(\frac{h}{r}\right) + 0.019 \cdot \left(\frac{h}{r}\right)^2 \quad C_4 = -3.572 - 0.634 \cdot \left(\frac{h}{r}\right) + 0.010 \cdot \left(\frac{h}{r}\right)^2$$

$$k_{\text{th}} = C_1 + C_2 \cdot \left(\frac{h}{D}\right) + C_3 \cdot \left(\frac{h}{D}\right)^2 + C_4 \cdot \left(\frac{h}{D}\right)^3 = 2.969$$

$$k_t = 1.11 \cdot k_{\text{th}} \cdot \left(-0.0159 + 0.2243 \cdot \frac{\alpha}{150} - 0.4293 \cdot \left(\frac{\alpha}{150}\right)^2 + 0.3609 \cdot \left(\frac{\alpha}{150}\right)^3\right) \cdot k_{\text{th}}^2 = 3.429$$

Calculate Moment and Stress

$$M = 20 \cdot \int_{\frac{d}{2}}^{\frac{2.5t}{2}} \rho \cdot (D + 0.2 \text{ in}) \cdot (t_{\text{notch}} + 2 \cdot l \cdot \tan(\theta)) \cdot l \cdot g \, dl = 0.871 \text{ J} \quad \sigma_{\text{nom}} = \frac{6 M}{t \cdot d^2} = (2.516 \cdot 10^8) \frac{\text{N}}{\text{m}^2}$$

$$\sigma_{\text{max}} = k_t \cdot \sigma_{\text{nom}} = 125.11 \text{ ksi}$$

$$\sigma_{\text{max}} = k_{\text{th}} \cdot \sigma_{\text{nom}} = 108.33 \text{ ksi} \quad \text{FS} = \frac{73.200 \text{ ksi}}{\sigma_{\text{max}}} = 0.676$$

After Recovery Results

- Beams CT scanned to check pore distribution and crack length in neck region
- 20g beams saw more deformation than 30g set but not total failure as expected
- Next Steps: Integrate strain gauges and qualify material

