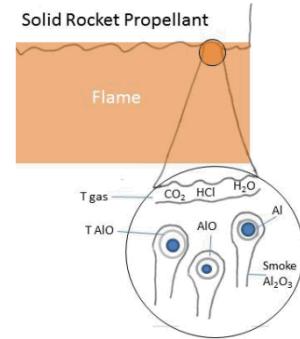
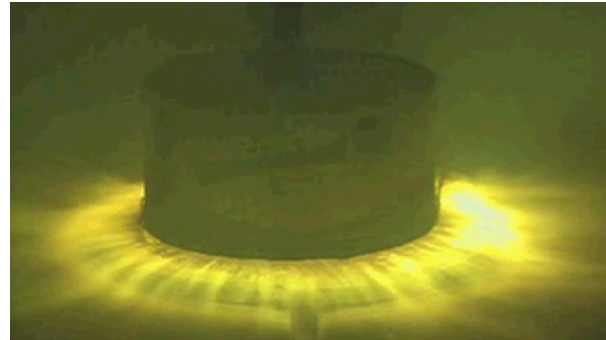
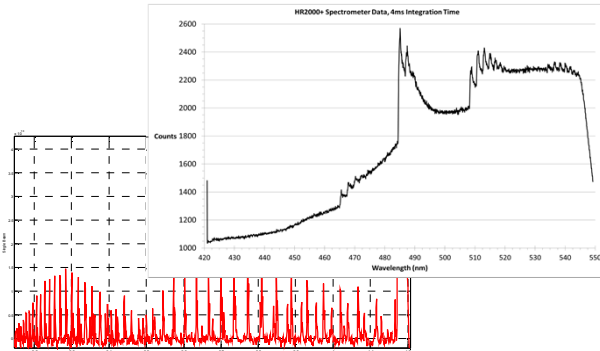


Exceptional service in the national interest



Measurement of solid rocket propellant exhaust gas temperatures using molecular spectroscopic methods

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David Surmick^b, Leland Sharp^c, Edward Bystrom^a and Aren Haug^c

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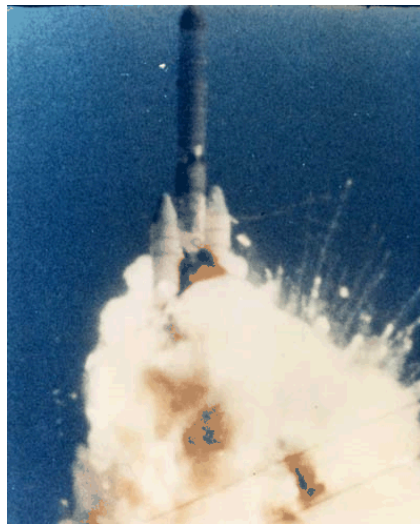
^bThe University of Tennessee Space Institute, Tullahoma, TN 37388

^cNew Mexico State University, 1780 E University Ave, Las Cruces, NM 88003

Introduction

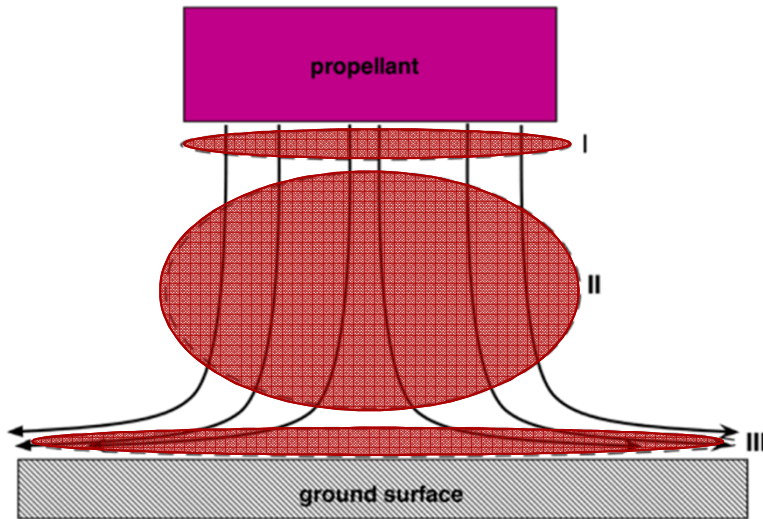
Program objectives

- Define the **thermal-chemical-physical** environment in and around a plume formed from a burning fragment of solid rocket fuel at atmospheric pressure.
- Use the specification to quantify the risk associated with a launch pad abort involving solid fueled rocket motors and high hazard payloads (e.g., deep space power source).



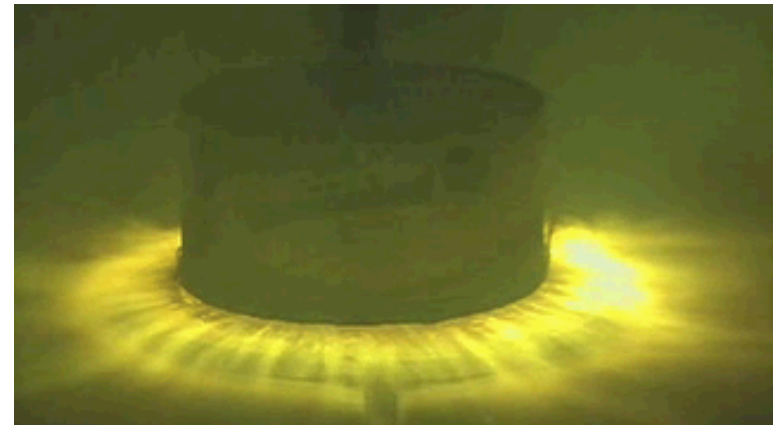
Titan 34D-9 accident sequence, 1986

Regions of Analysis

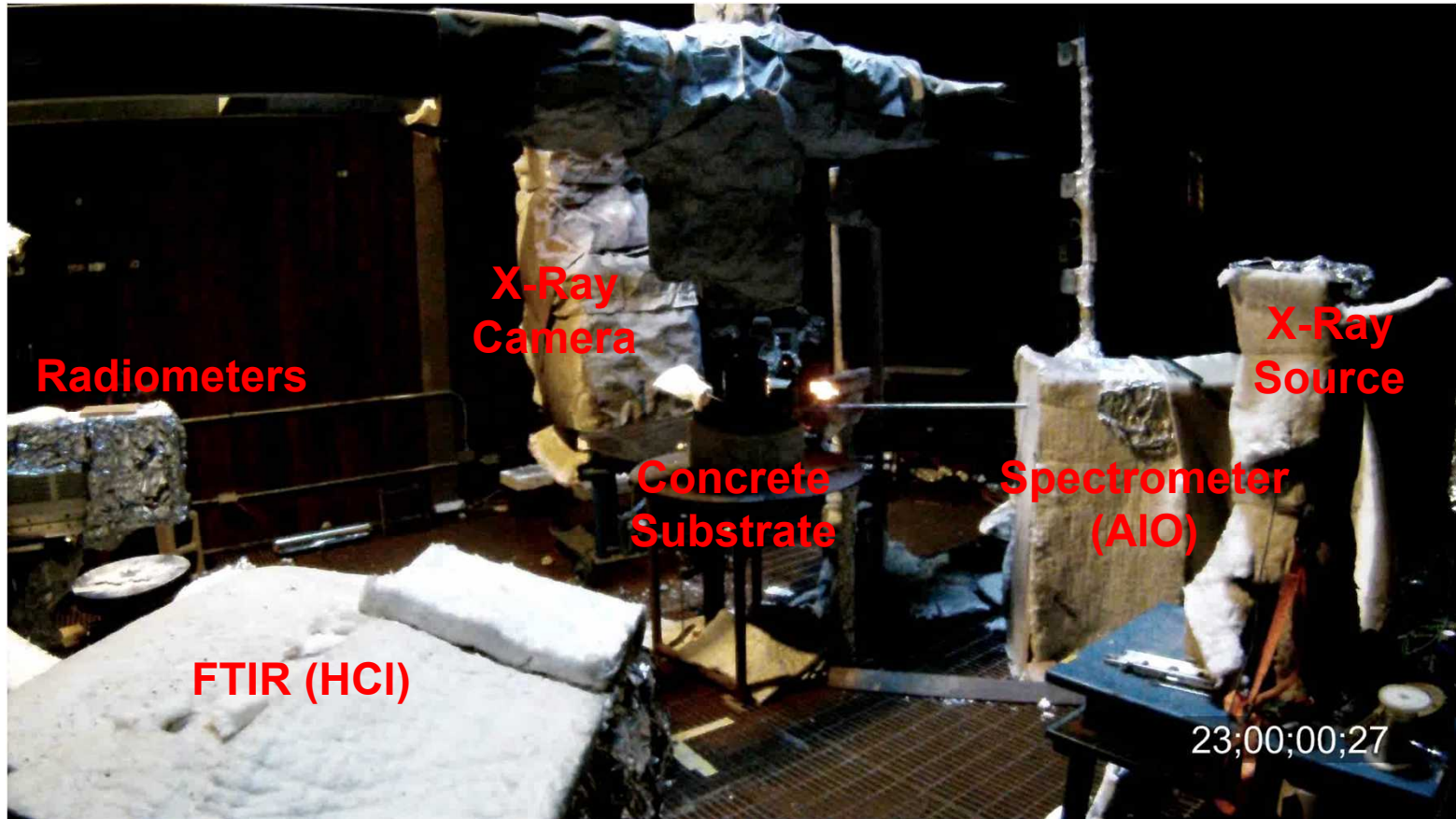


- Region I: combustion of Al + ammonium perchlorate (+ binder, etc.)
- Region II: populated by hot, emitting gases (HCl, CO₂, H₂O) and particles (AlO, Al₂O₃, etc.)
- Region III: interaction with hazardous payload; possible feedback to Region II (e.g., re-radiation)

- Physico-chemical behavior of burning Al strongly correlated to the environment, *i.e.*, properly functioning motor at high pressure/well defined geometry *versus* off-normal occurrence.
- **Major concern:** vaporization of hazardous payload material in Region III, condensation on available particulate and transport to the environment.



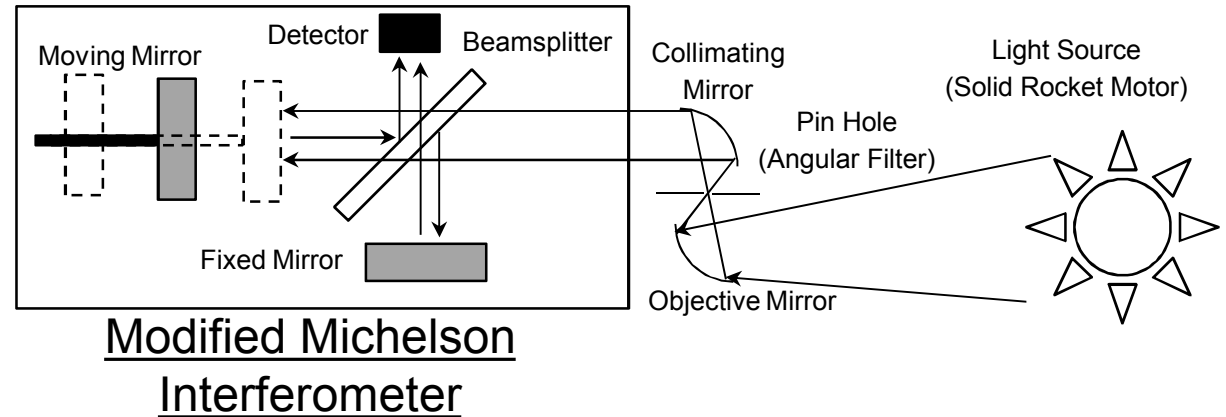
Test Case – March 12, 2014



Plume from burning propellant is complex:
highly emitting
multiple chemical species and temperature-regions
particulates / smoke
substrate degradation / interaction

HCl Measurements

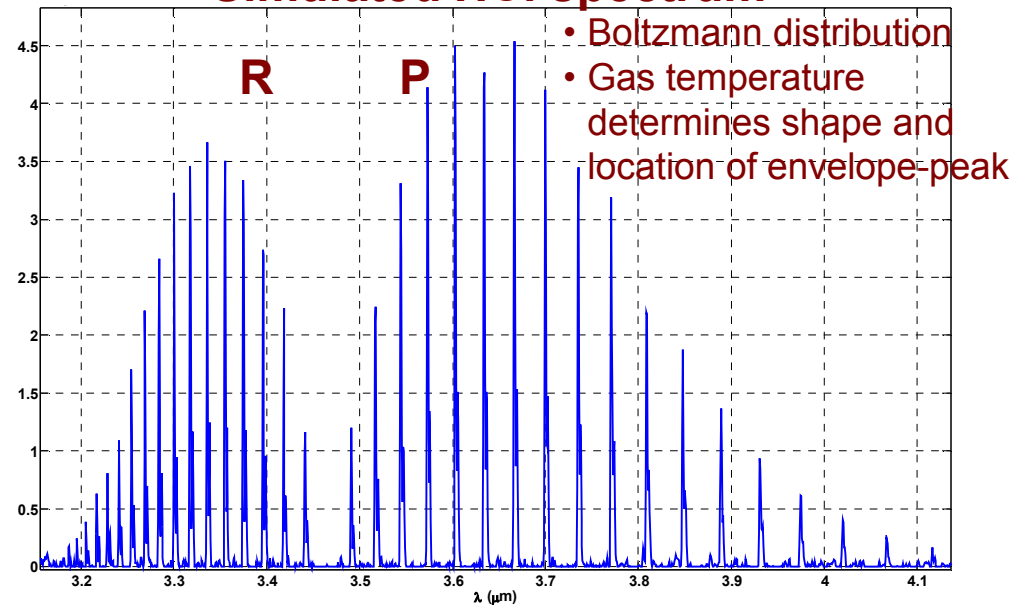
Fourier Transform IR Interferometry



HCl considered target of opportunity to measure gas temperature

- ✓ High S/N for the HCl spectra
- ✓ Spectral regions with minimal overlap
- × Noisy data captured on loss of sight of the flame due to:
 - Propellant inhibitor failure
 - Extremely small gaps
 - High particle-density

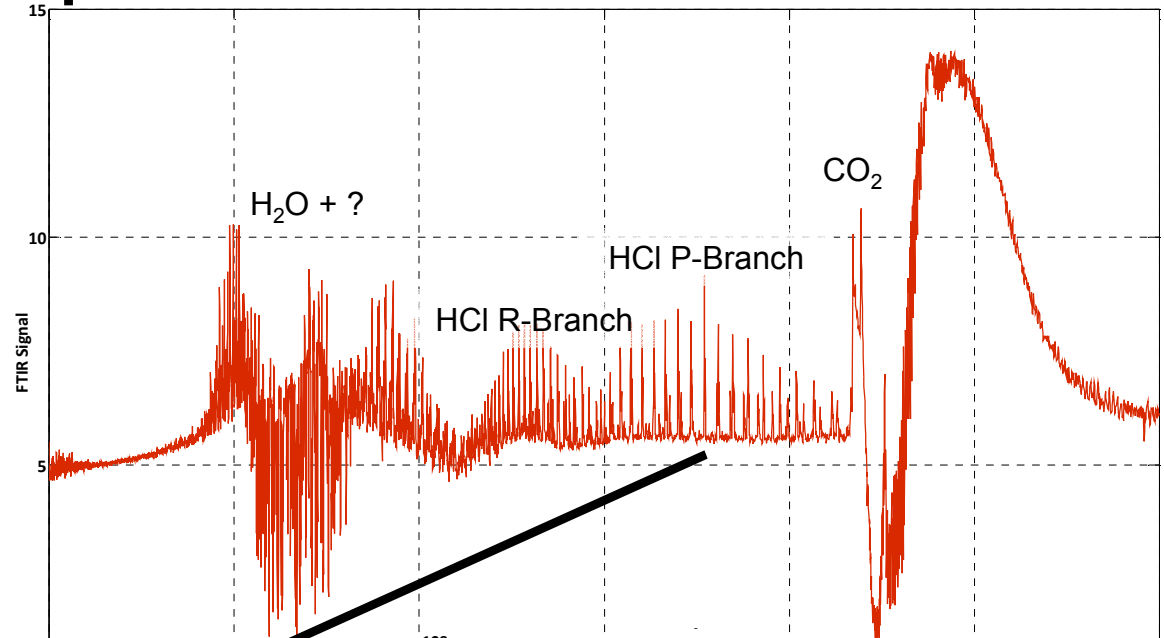
Simulated HCl spectrum



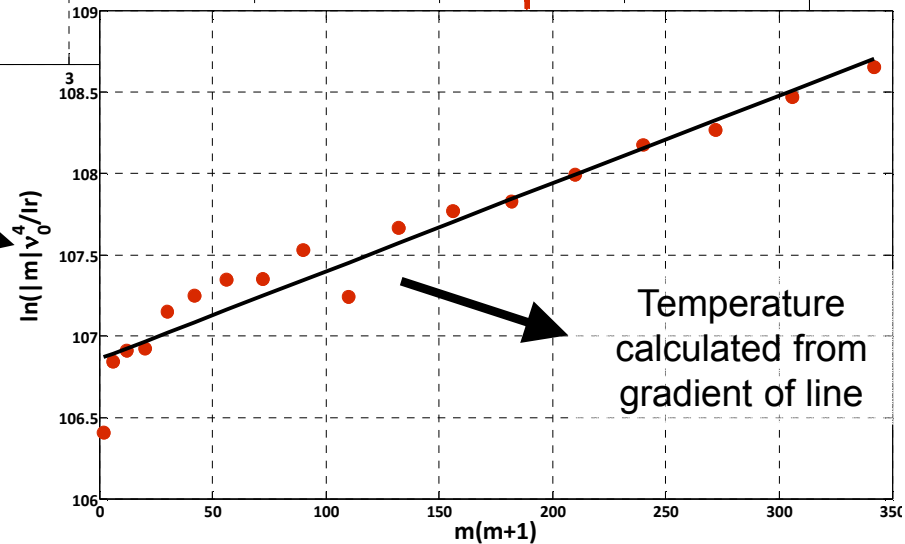
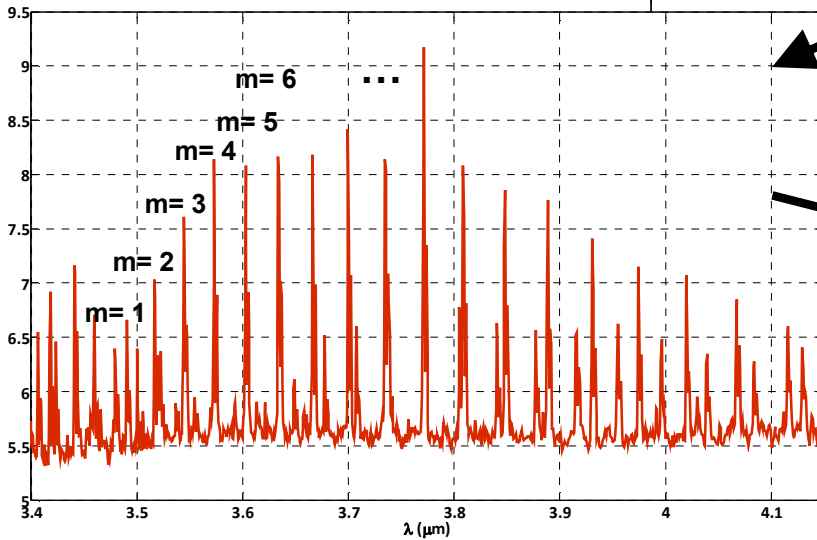
HCl Measurements

Spectrum Measured

- Devcon Inhibitor
- 6" Diameter
- 1" Gap
- Preliminary temperatures calculated using Boltzmann curves

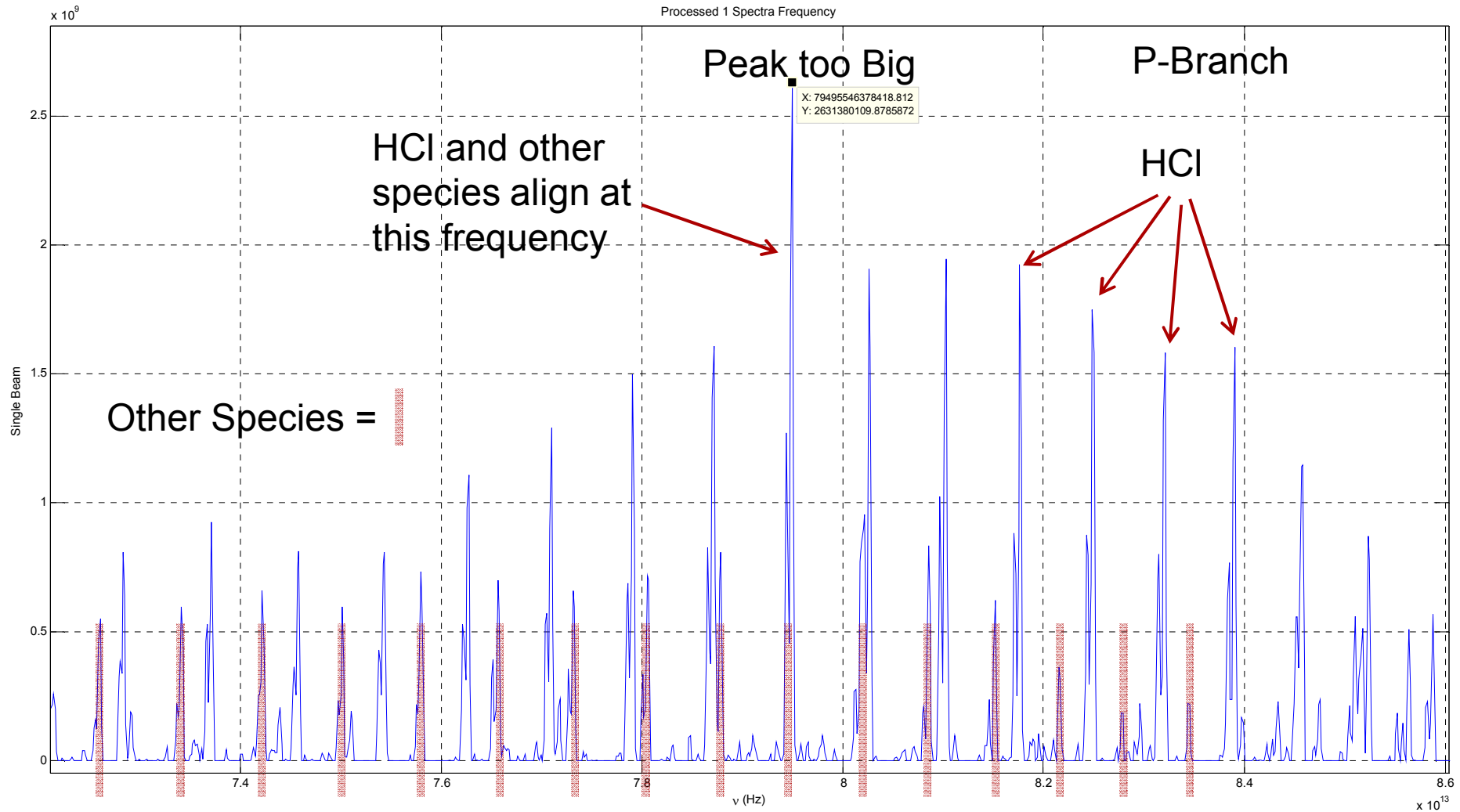


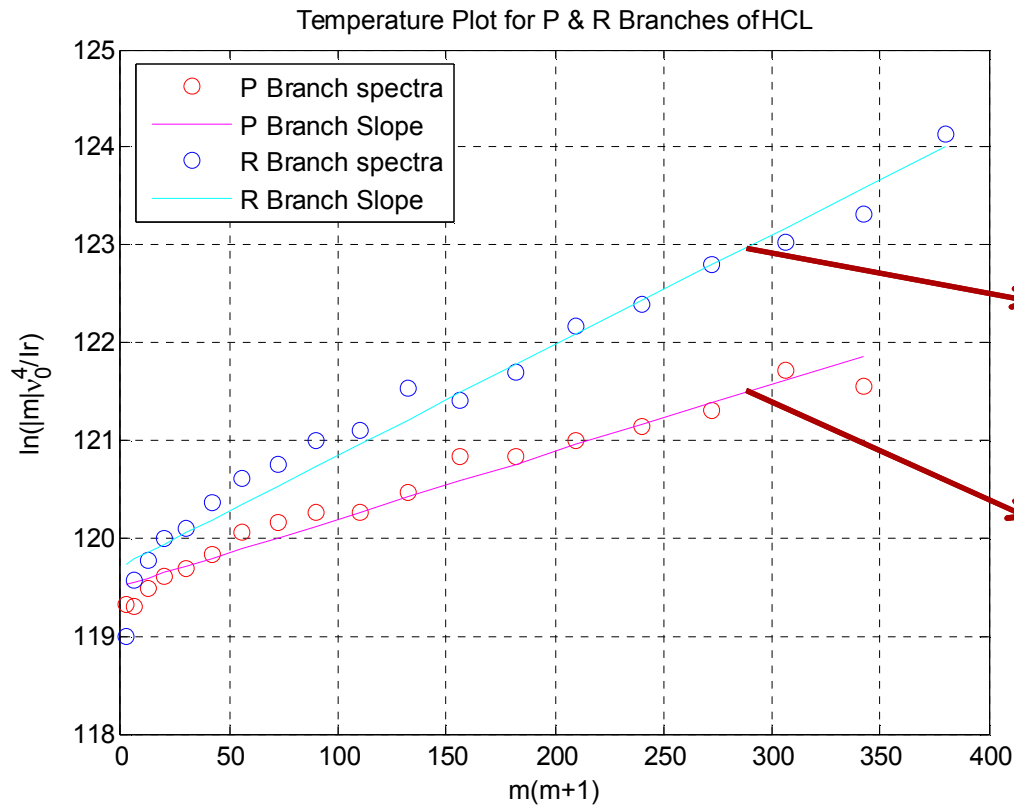
HCl P-Branch



HCl Measurements

Unexpected Species, What is it?





HCl temperature ~ 1200 – 1500 K
- too low?

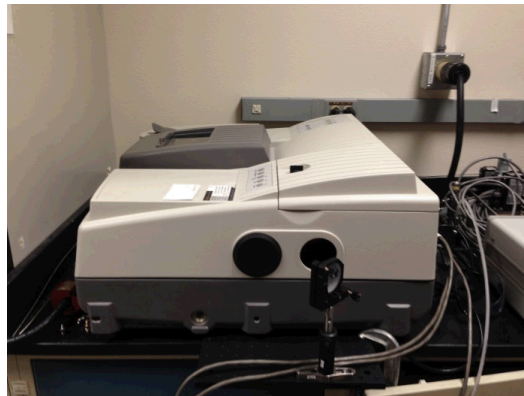
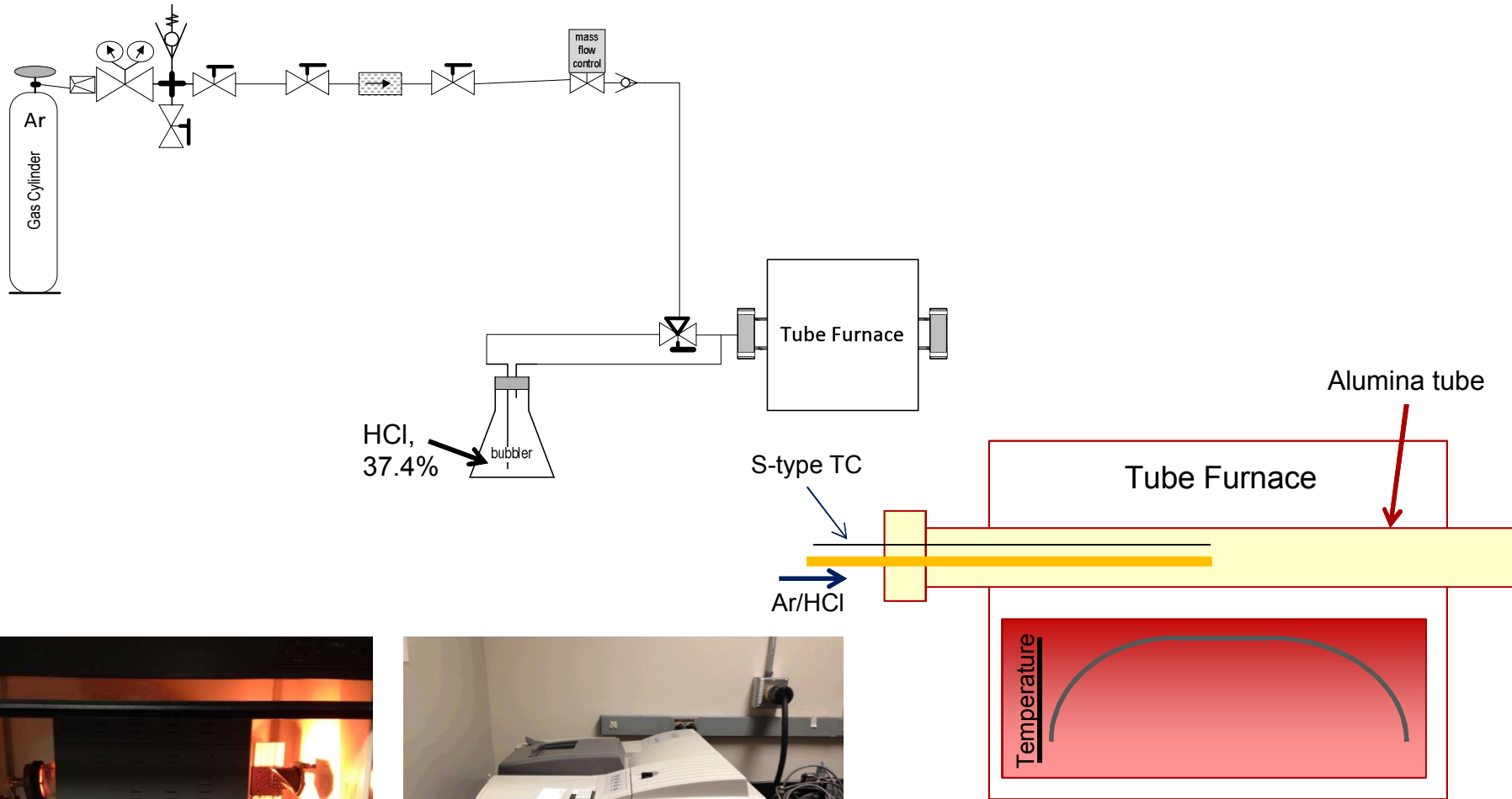
HCl temperature ~ 3000 K
- not un-reasonable

- Difference between P- and R-branch results
- Unknown secondary fingerprint

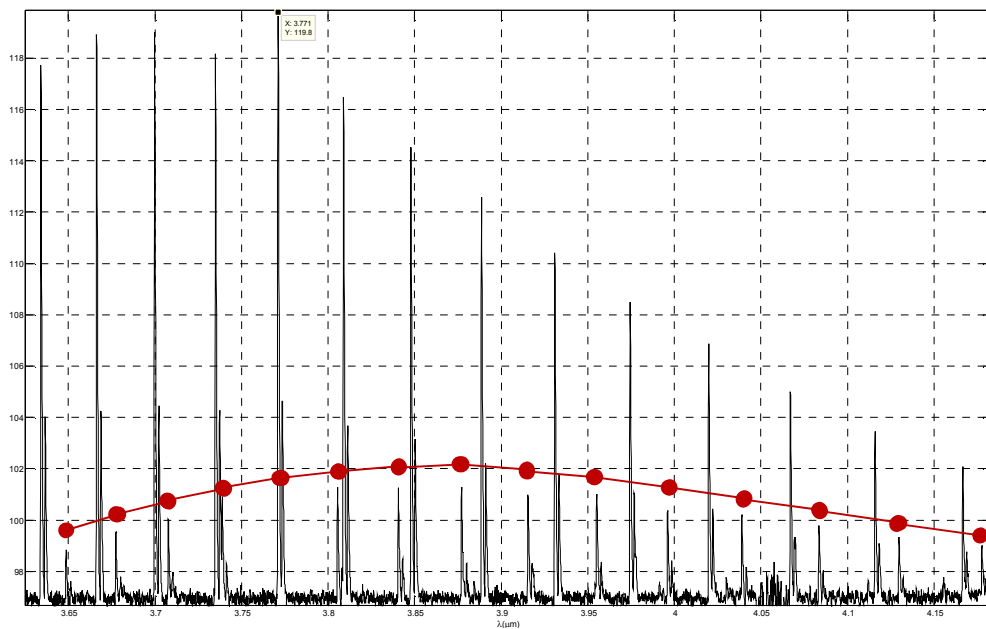
→ **Need for experiments under controlled conditions**

Calibration & Validation

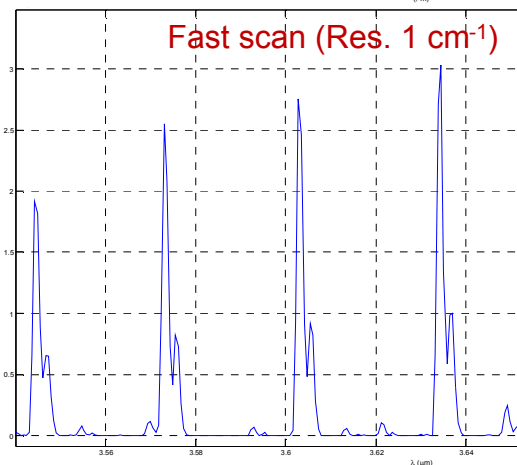
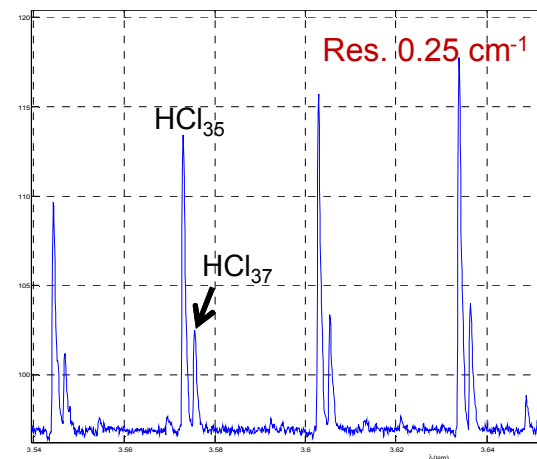
“clean” environment, controlled temperature



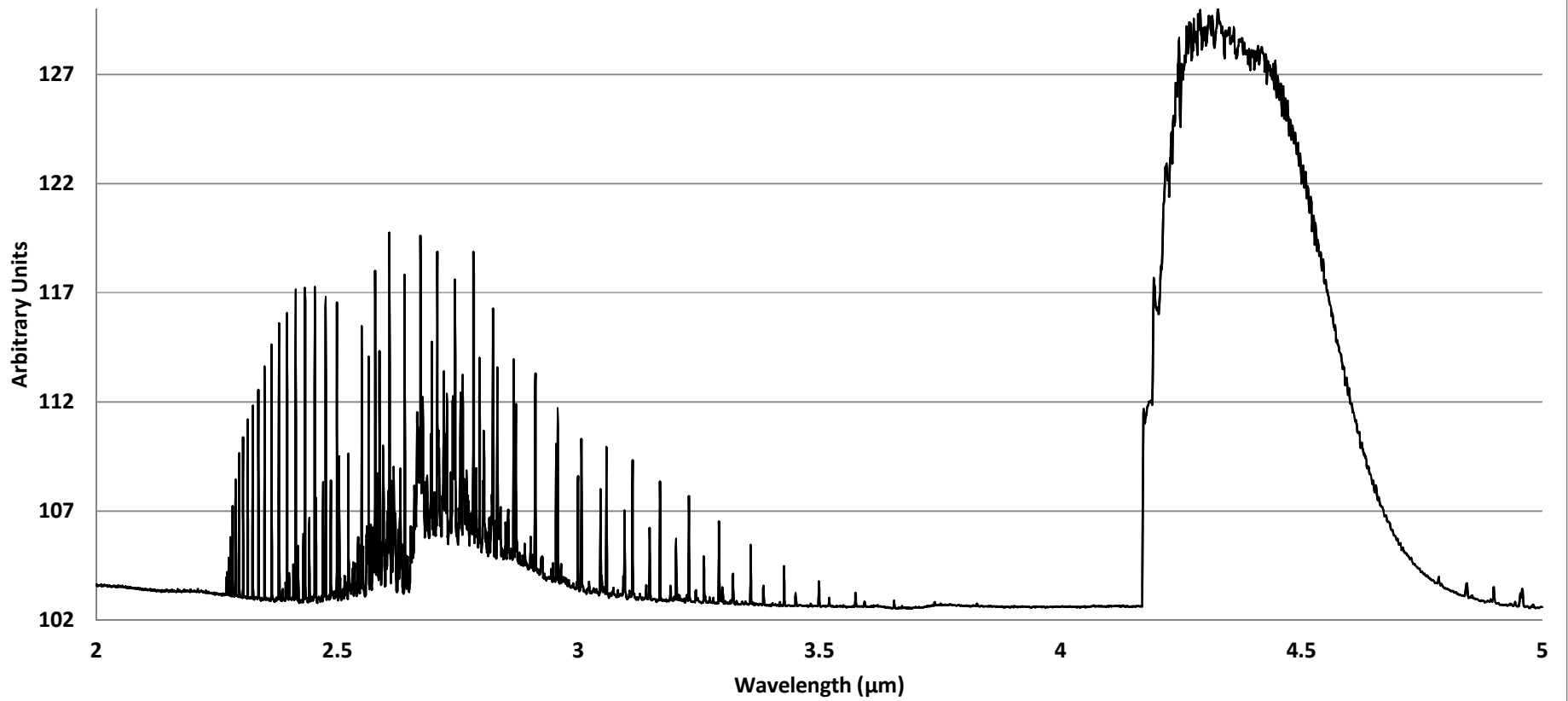
Tube furnace operations

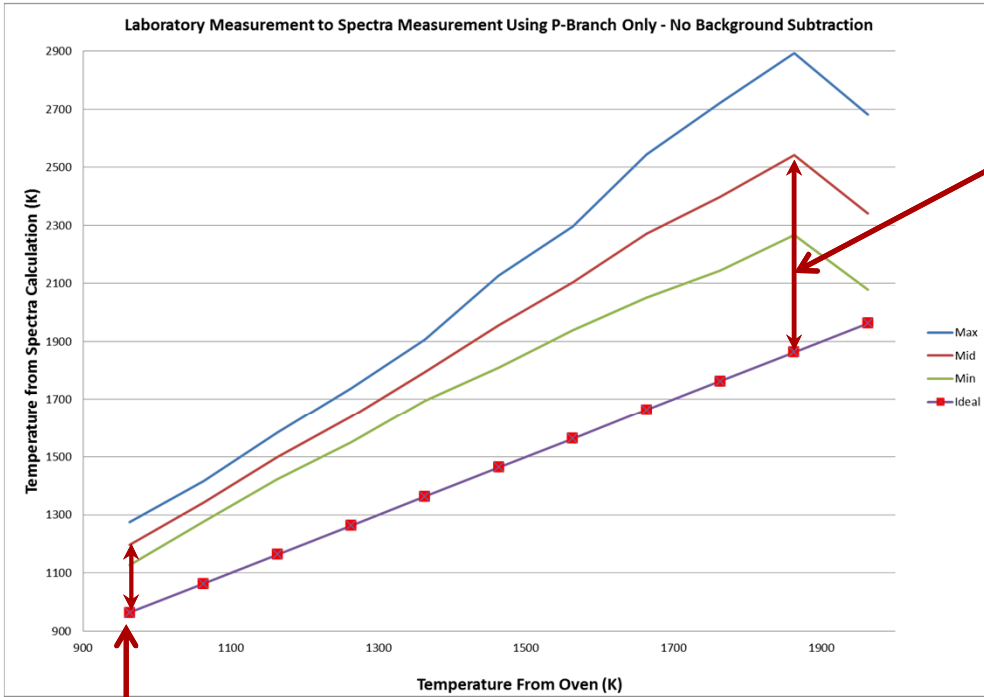


- Even under “clean” conditions, unknown secondary pattern persists
- Isotope effect resolvable
- At higher resolution, could probably resolve overlap at 3.77 μm



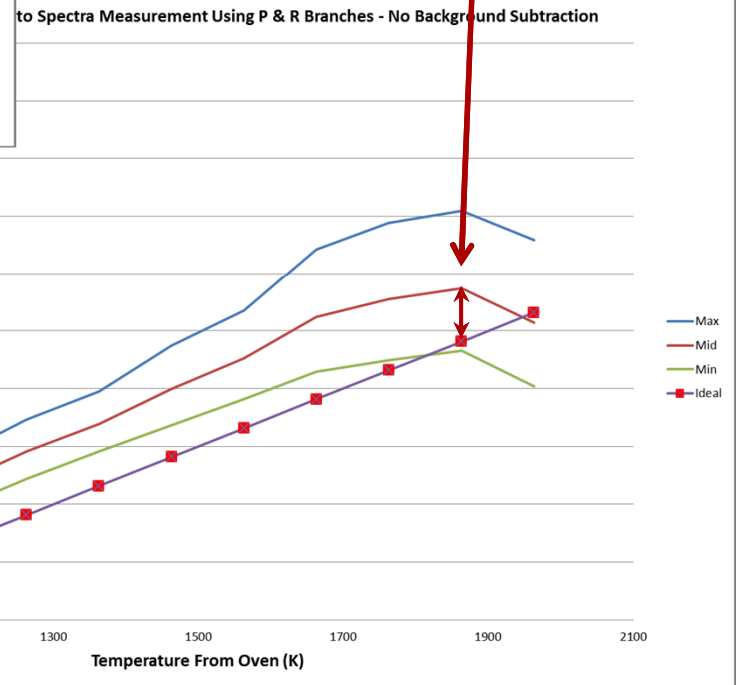
H₂O at 1675K





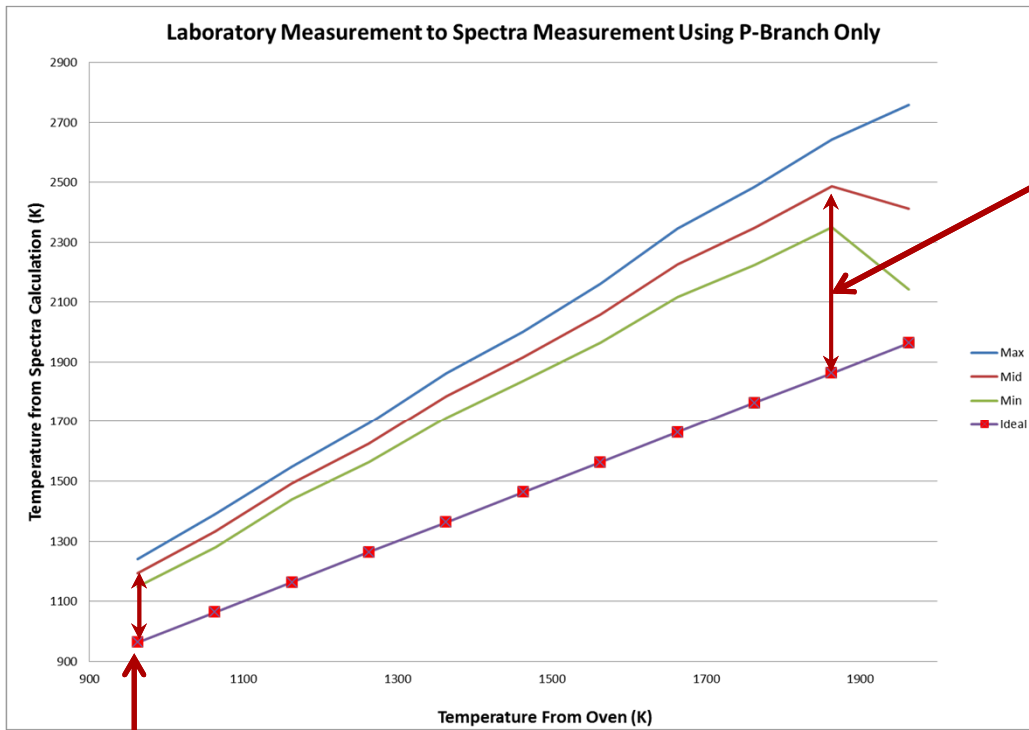
$\Delta = 233 \text{ K}$

$\Delta = 680 \text{ K}$



$\Delta = \sim 200 \text{ K}$

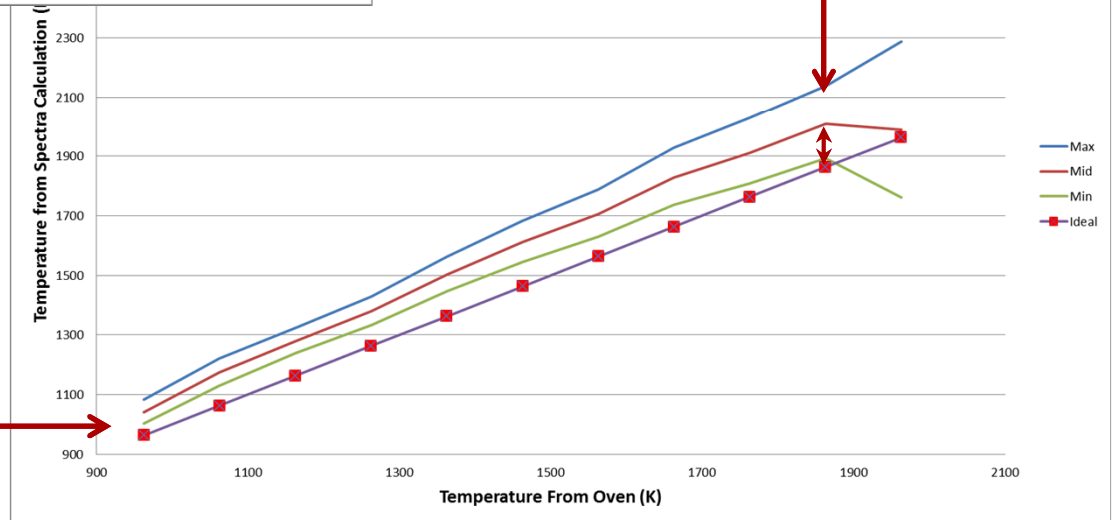
$\Delta = \sim 200 \text{ K}$



$\Delta = 230 \text{ K}$

$\Delta = 630 \text{ K}$

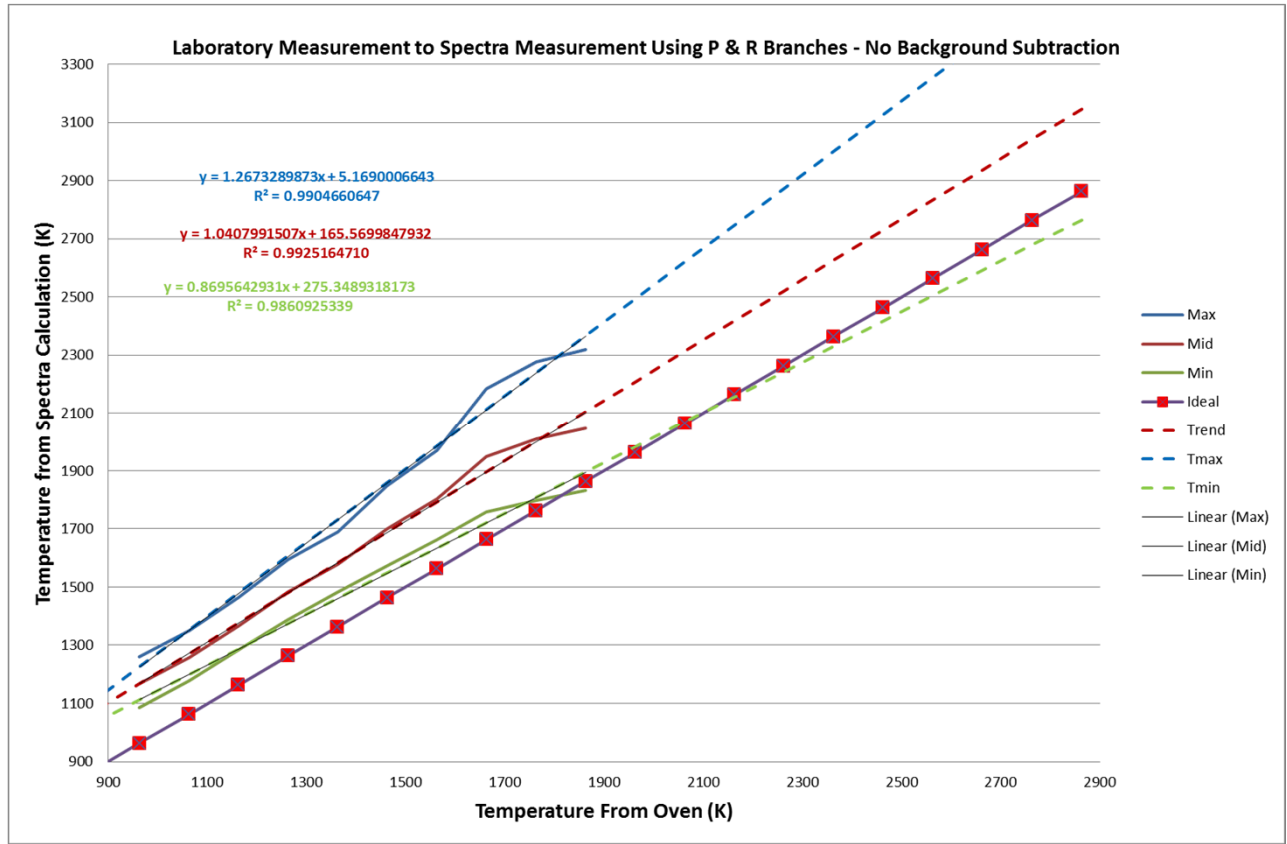
Measurement to Spectra Measurement Using P & R Branches



$\Delta = 145 \text{ K}$

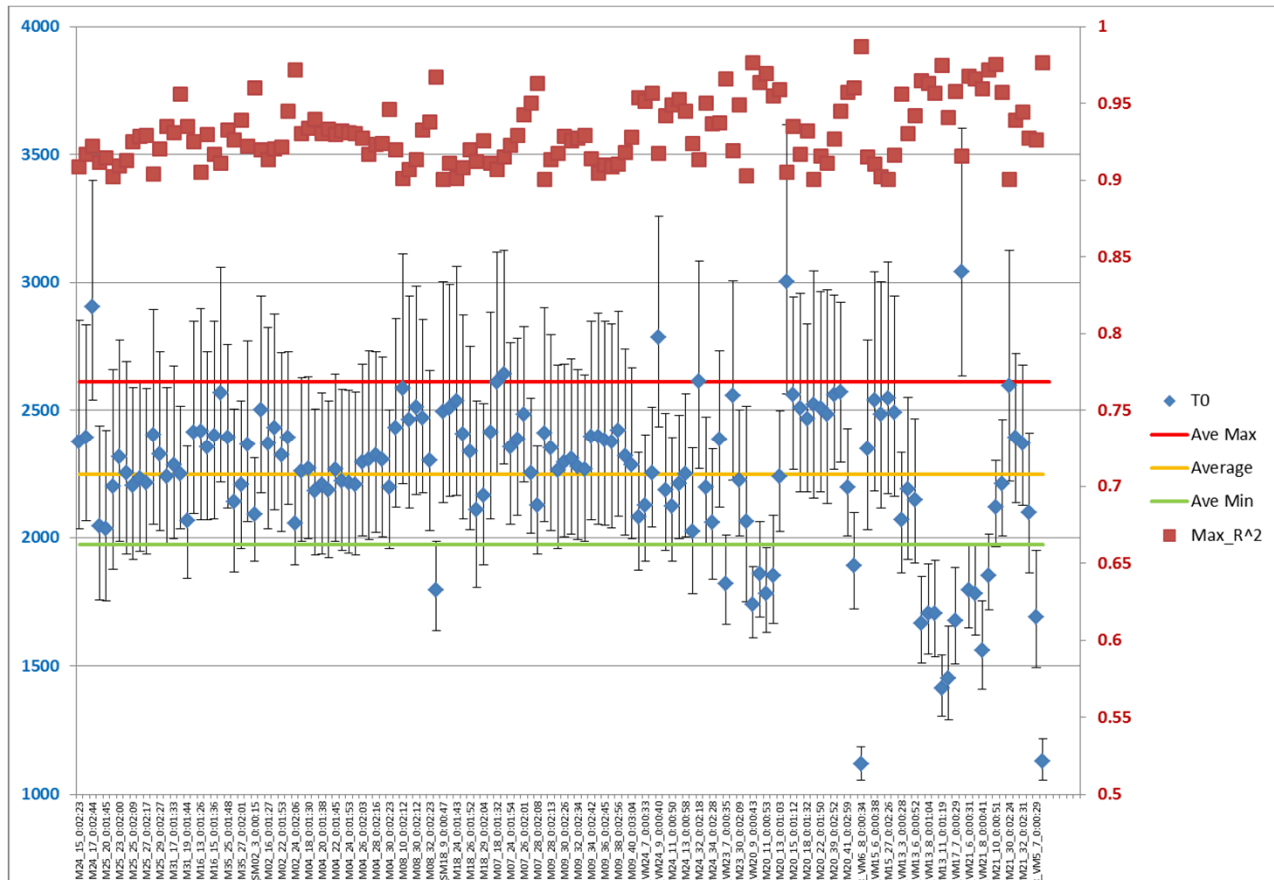
$\Delta = 78 \text{ K}$

Laboratory Experiment: Correction P & R



- Linearizing the trend (red lines) The error of average of P & R branches is **176K to 282K** for calculated temperatures with HCl spectrum of **263K to 2863K**

$$T_{HCl_{Final}} = T_{HCL} - AT_{HCL} - B, \text{ where } A=0.0391998309 \text{ and } B=159.0796693885$$



Temperature for tests on two different types of propellants (blue diamonds – left ordinate). The reported temperatures have coefficient of determination (r^2 – red squares – right ordinate) larger than 0.9. The temperatures were calculated using an approximation to the Boltzmann equation from the R & P branches.

HCl Measurements

Summary

HCl is an opportunistic target for temperature measurement in propellant fires

- component of propellant plume
- strong optical emission
- relatively free from spectral overlap
- simple FTIR setup works fine

Still work to be done to refine model, based on tube furnace tests

Gas temperature in typical propellant plume between 2000 and 2600 K

Thank You

Funding for this work:



National Aeronautics and
Space Administration

Jet Propulsion Laboratory
California Institute of Technology
Pasadena, California